IN THE SPECIFICATION

Please replace the paragraph beginning at page 5, line 5, with the following rewritten paragraph:

~~Figures 3A, 3B, and 3C show~~ Figure 3 illustrates a split ring, a laminar ring, and a multiple split ring, ~~respectively,~~ that can be used in a support structure of the present invention.

Please replace the paragraphs beginning at page 6, line 25, and ending at page 8, line 15, with the following rewritten paragraphs:

~~In another embodiment of a combustion chamber, as illustrated in Figure 2,~~ In a further embodiment, combustor chamber 30 ~~has a~~ may have an annular shape having common longitudinal axis 20 with turbine engine casing 50. The term "annular shape" includes structures that have constant or varying diameter along its longitudinal axis. Turbine engine combustion chamber section 10 has an outer metal casing 50 and an inner metal casing 70 that are made of a metal or an alloy as disclosed above. The inner surface 52 of outer metal casing 50 and the outer surface 72 of inner metal casing 70 may be desirably coated with a conventional ceramic insulating thermal barrier coating material such as oxides of yttrium and zirconium or oxides of zirconium and scandium. The thermal barrier coating reduces the heat transmission to the metal casing and, thereby, lowers its average temperature. An outer combustor liner 100 is disposed adjacent to inner surface 52 of outer metal casing 50. An inner combustor liner 120 is disposed adjacent to outer surface 72 of inner metal casing 70. Combustor liners 100 and 120 are disposed in combustor section 10 such that they can move independently relative to casings 50 and 70 to accommodate the different coefficients of thermal expansion of the materials of the combustor liners and of the casings. Combustor liners 100 and 120

can comprise ceramic matrix composites, such as a silicon carbide matrix reinforced with silicon carbide fibers, as disclosed above.

A support or sealing structure of the present invention is ~~also~~ shown in ~~Figure 1,~~ Figure 2, which is not drawn to scale. The structure comprises a support member 200 for attaching combustor liner 100 to combustor casing 50 and still allows for an independent radial or axial movement of combustor liner 100 relative to combustor casing 50. In an embodiment wherein the combustor has an annular shape, such as that illustrated in ~~Figure 2,~~ Figure 1, similar structures (not shown) of the present invention also are employed to attach inner combustor liner 120 to inner combustor casing 70. In one embodiment of the present invention, support member 200 generally has a cylindrical annular shape, the outer circumferential surface of which is fixedly attached to inner surface 52 of outer casing 50. Support member 200 has a circumferential portion 202 and two opposed radially inward extending portions 204. Circumferential portion 202 and radially inward extending portions 204 together define a cavity 206. Two mating attachment structures comprising laminar or split rings 208 and 210 are disposed within cavity 206. The first attachment structure comprises a plurality of first laminar or split rings 208. Rings 208 are disposed adjacent to an inner circumferential surface of cavity 206 and are spaced apart from one another and from outer surface 102 of outer combustor liner 100. A plurality of second laminar or split rings 210 having an average diameter generally smaller than that of first rings 208 and being disposed adjacent to outer surface 102 of combustor liner 100. Second rings 210 are spaced apart from one another such that each of second rings 210 slidably engages between and overlaps two adjacent first rings 208. The outer circumferential surface of second rings 210 is spaced apart from the inner surface of circumferential portion 202 of cavity 206. Laminar or split rings 208 and 210 are capable of expanding or contracting in the radial direction so as to accommodate the different coefficients or rates of thermal expansion or contraction of combustor liner 100 and casing 50. Each of first and second rings 208 and 210 may have a configuration of a single split ring, a double laminar ring, or a plurality of split rings having the same

diameter and attached together ~~as shown in Figures 3A, 3B, and 3C~~, illustrated and referred to as A, B and C respectively in Figure 3. Laminar or split rings 208 and 210 can be made of a ceramic material such as a ceramic matrix composite material as disclosed above. However, suitable metals with or without a thermal barrier coating also may be used for rings 208 and 210. As combustor liner 100 is exposed to a very high temperature of the combustion product gas, it tends to expand in the radial direction. Second rings 210 expand with combustor liner 100 because of the existence of a split in or the discontinuous nature of the rings. Each of second rings 210 thus slides over at least an adjacent first ring 208 to accommodate the expansion of combustor liner 100 so as not to allow a stress build-up therein.